

GENERAL DYNAMICS F-16A FIGHTER

The F-16 was developed by General Dynamics Corporation on the Lightweight Fighter program directed by the U.S. Air Force. Its protytype, designated YF-16 (serial No. 72-01567) made the first official flight 2 February 1974 and recorded a speed of Mach 2 on 11 March 1974. The second YF-16 (72-01568) flew for the first time on 9 May 1974. After a long term of ACF (the Air Combat Fighter) evaluation on the F.16 and its competitor Northrop YF-17, the adoption of the F-16 was decided. In April 1975, a contract awarded to General Dynamics for eight engineering development aircraft, comprising six single-seat F-16As and two two-seat F-16Bs. These F16As/Bs were a little bit increased in the size and weight as compared with the YF-16; the winc area was enlarged from 26.01m² to 27.87m² and the wheel base prolonged from 3.73 m to 4.00 m to improve their operational serviceability.

The first F-16A (75-745) made its maden flight on 8 December 1976 and the F-16B, on 8 August 1977. After evaluation the decision was made by the U.S. Air Force to adopt 1,388 F-16As/Bs, following 583 airframes by the six nations of Belgium, Denmark, the Netherlands, Norway, Iran and Israel; in total, 1,971 aircraft production was determined as of February 1978.

Co-production arrangement has been made with more than thirty companies in the West European countries, including Fokker in the Netherlands and SABCA in Belgium, so that the F-16s are expected to be an international production type fighter plane to replace the F-104s in current service.

The F-16, often referred to as the Zero Fighter of today, is indeed an epockmaking fighter/bomber, though small in size and light in weight but highly efficient with the features as follows:

- 1. Newly designed for highter performance and productivity; it is much easier to manufacture, service and operate as compared with other Mach-2 class fighters.
- 2. The intersection of the body/wing is faired with no boundary line to form a configuration that will reduce parasite drag. This "blended wing/body", designed in anticipation of the future airplanes, is wing/body", designed in anticipation of the lattire arrhanes, is provided with strakes extending along the fuselage forebody and speed brakes at the rear ends of their extension, which greatly increase climb, dive and land/take off performance.

 3. The seat is inclined 30° aft and the hee is raised to render the pilot to
- bear up against the fierce movement and to control the plane even in the heavily-loaded condition.
- 4. The control column is of right hand small toggle type, unlike the front-central stick type on the conventional fighter planes. It is the handle of fly-by-wire electronic control system, through which flight control is made by the precision self-controlled servo motor; the slats and the flaps are self-actuated in accordance with the rate of speed, and also danger is automatically detected and prevented.

The bubble canopy provides all-round vision, especially developing in the forward.

5. Though it is a lightweight monoplane, it fullfils all-weather mission with the newly designed auto-pilot/attack system. It is also possible to mount anti-air/support system without difficulty.

- 6. The power plant is the F100 turbofan engine that is light in weight but generates a large thrust. It is more economical in the fuel consumption, more silient in the noise and produces less exhaust gas than other engines on current fighters.
- 7. The air intake of the engine protrudes to the fore of nosewheels and machine guns not to absorb sand and dust at taxing, and exhaust gas at take off.
- 8. As for the armament, it is fitted with one M61A-1 20 mm multi-barrel cannon in the portside of the fuselage (500 rounds of ammunition, rapidity of fire: 6,000 rounds per minute), and has hardpoints at wingtips, on underwings and underfuselage, making nine weapon stations in total, to carry sidewinder, sparrow and other types of air-to-air missiles, bombs and the various kinds of weapons, or otherwise, to mount additional fuel tanks for a long distance flight. Total possible weapon load, with reduced internal fuel, is 6,894 kg, and 4,763 kg, with full internal fuel, which can well stand comparison with the large twin-engined fighter such as the F-15.
- 9. The design load factor of the F-16 is 9G at take-off weight in aircombat mission. This value is high as compared with those of other current fighters and due enough to render the F-16 to execute any manoeuvre.
- 10. To cut production costs, the F-16 has a structure which requires no special materials nor extra manufacturing machines and tools. The fuselage structure consists of approx. 83% of aluminum alloy, 4.7%steel and approx. 4.2% composite materials. This keeps material cost to a minimum and facilitates co-production in foreign countries. As mentioned above, so many characteristics as never seen on the conventional fighters are incorporated in the F-16 and blended into a splen-

did configuration and superb performance of its own. The F-16B, the two-seat fighter/training airplane converted from the F-16A, has the same length of the F16A, but its engine output is decreased by approx. 17% and the internal fuel tankage reduced by 17%.

GENERAL DATA ON F-16A

Power Plant

: Pratt & Whitney F100PW-100 (3) turbofan engine with afterburner Thrust 8,165 kg

Fuel Capacity

: 4,010 lit. (interl tank) + 563 lit. (underfuselage tank) + 1,400 lit. (underwing tank) x 2 : Wing span 10.01 m (over missiles) Length overall

Dimensions

14.52 m Height overall 5.01 m

Weight & Loading: Weight empty 6,607 kg Weight full-loaded 10,205 kg Performance: Max. speed Mach 1.95 (at 11,000 m altitude, sidewinder x 2), Mach 1.2 (at sea level) Service seiling more than 15,240 m Max. rate of climb 12,802 m/min. (low-level Mach 0.7 with Mk82 bombs) Radius of action 925 kg (with no external armament), 547 kg (with six Mk82 bombs) Cruising range 3,705 km (with fuel tank, in ferry condition) T-O run 533 m (with 1,814 kg external load) Landing run 808 m (with 1,814 kg external load)

Manufactured in Japan & Packed in England by A. A. Hales Ltd., Hinckley, Leics.

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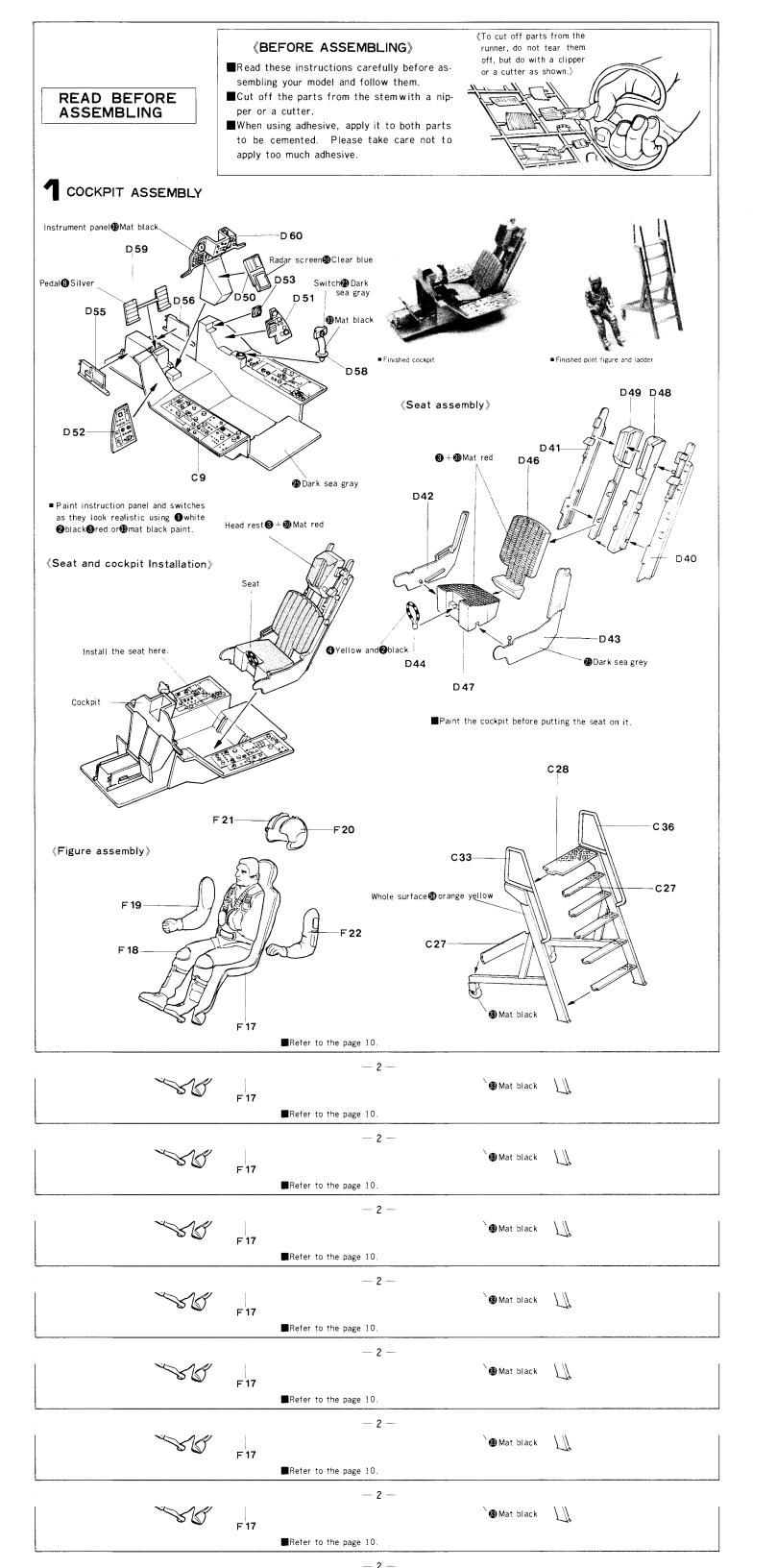
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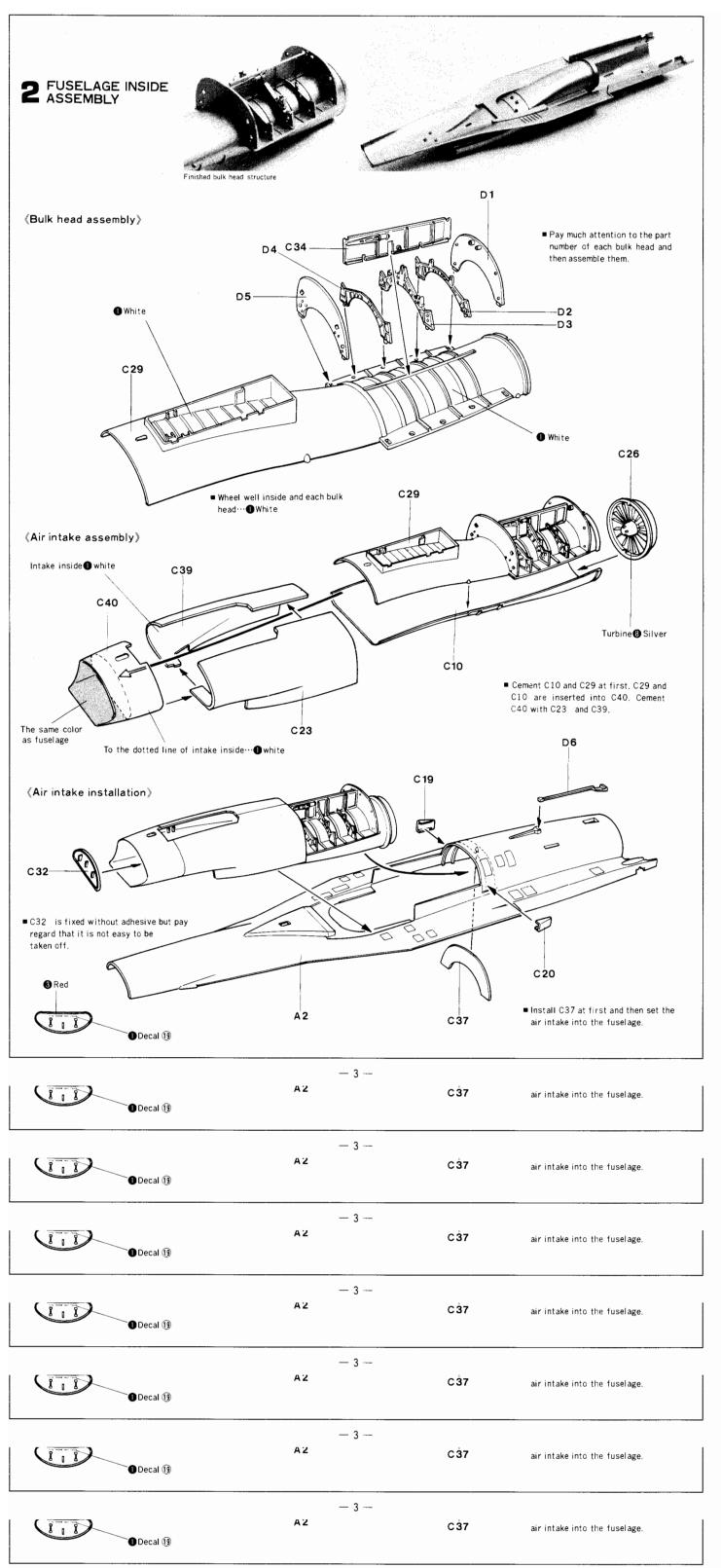
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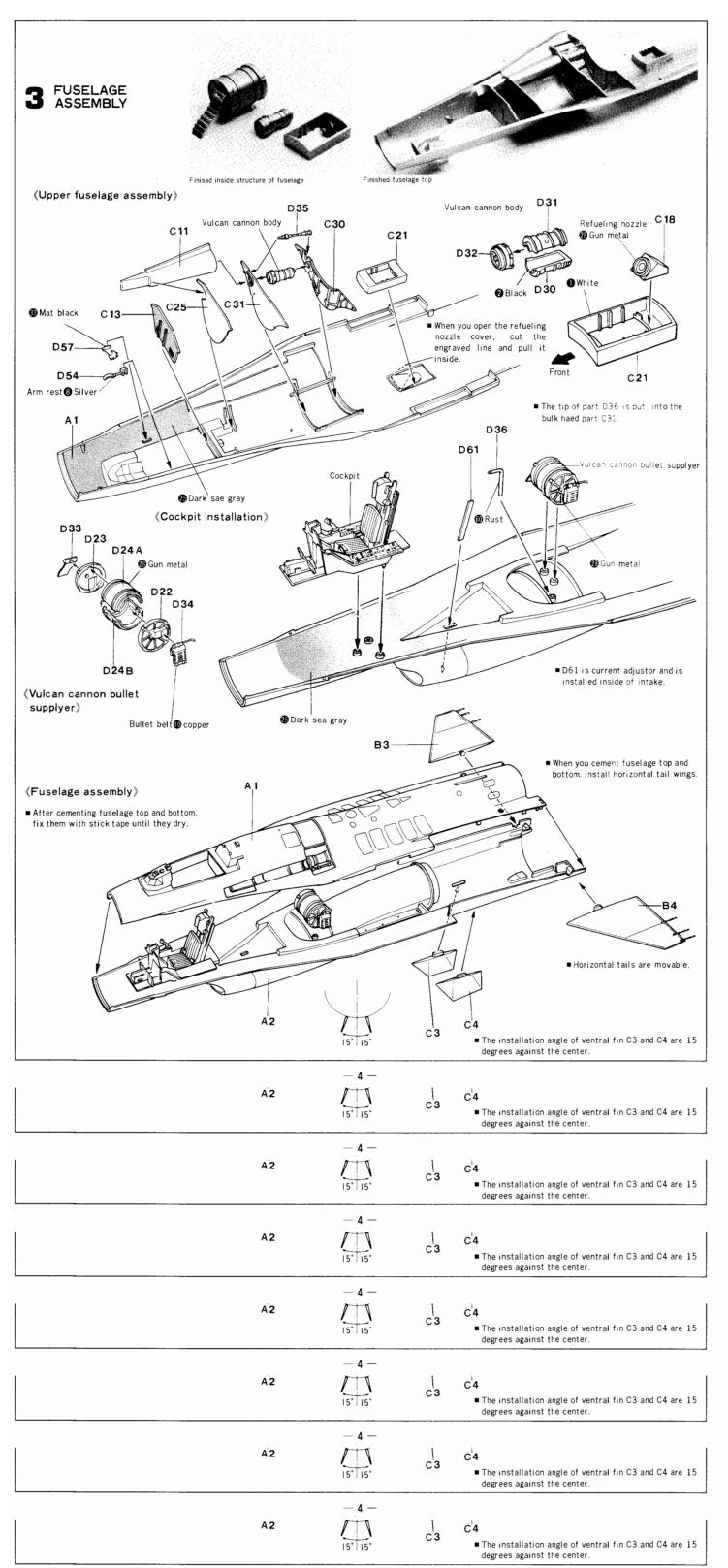
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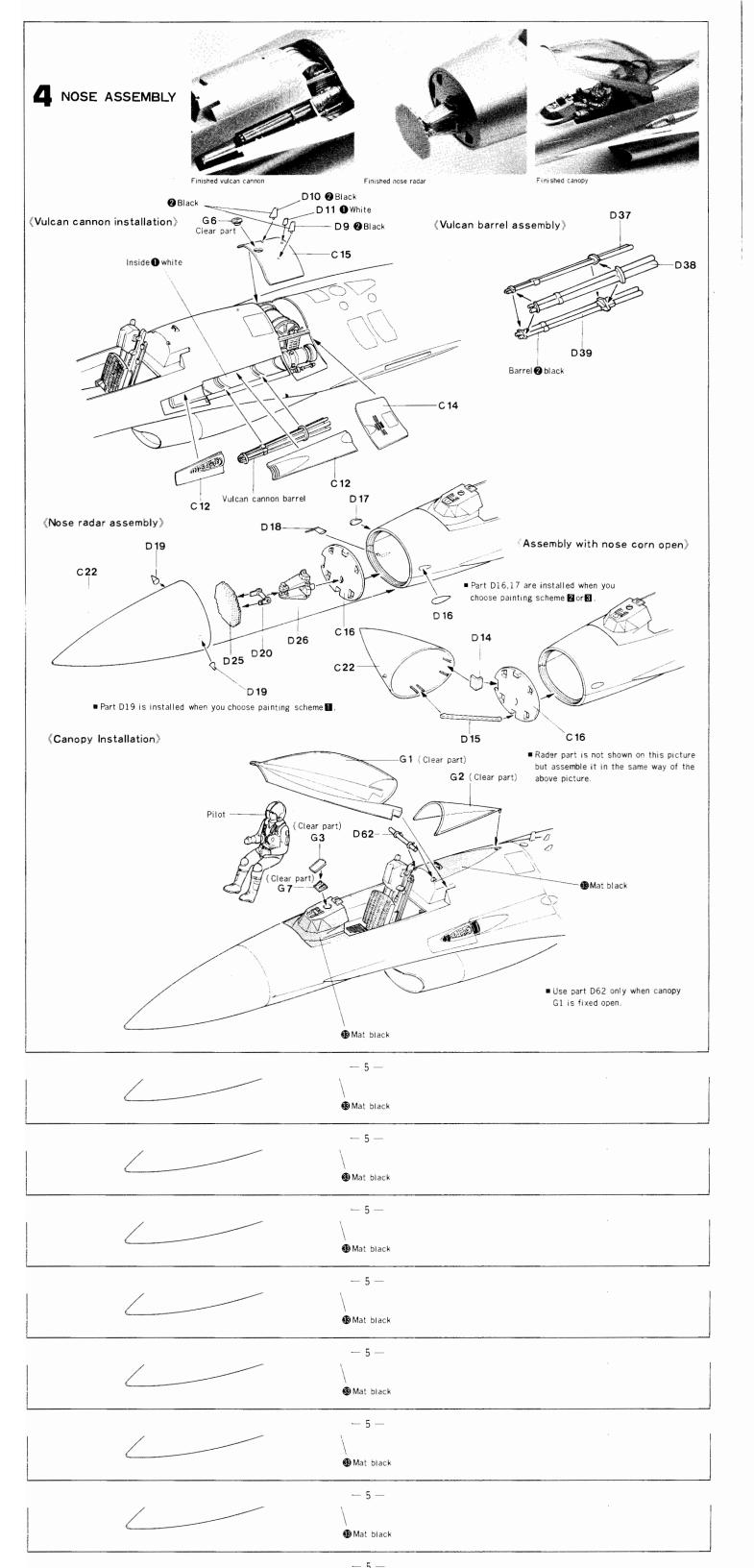
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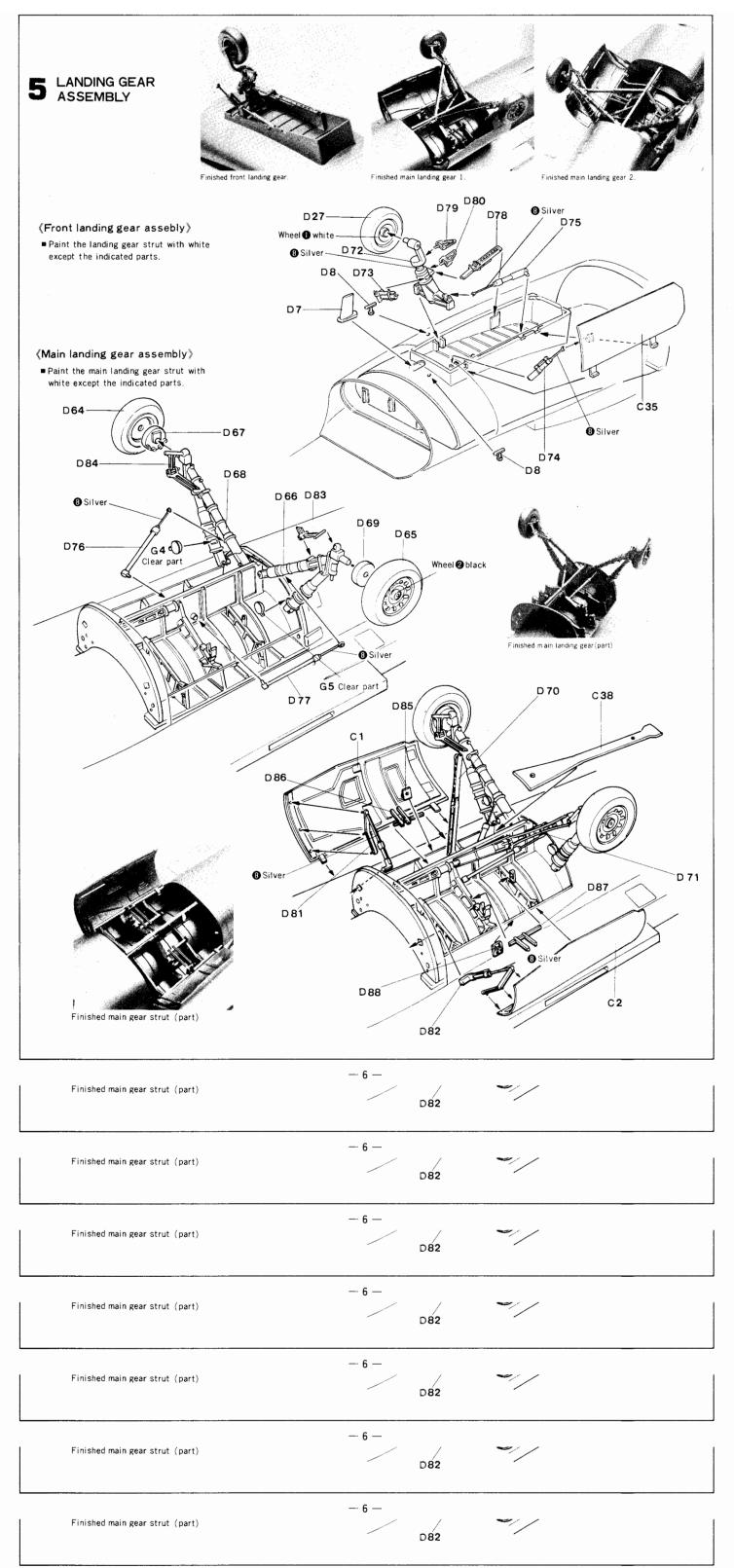
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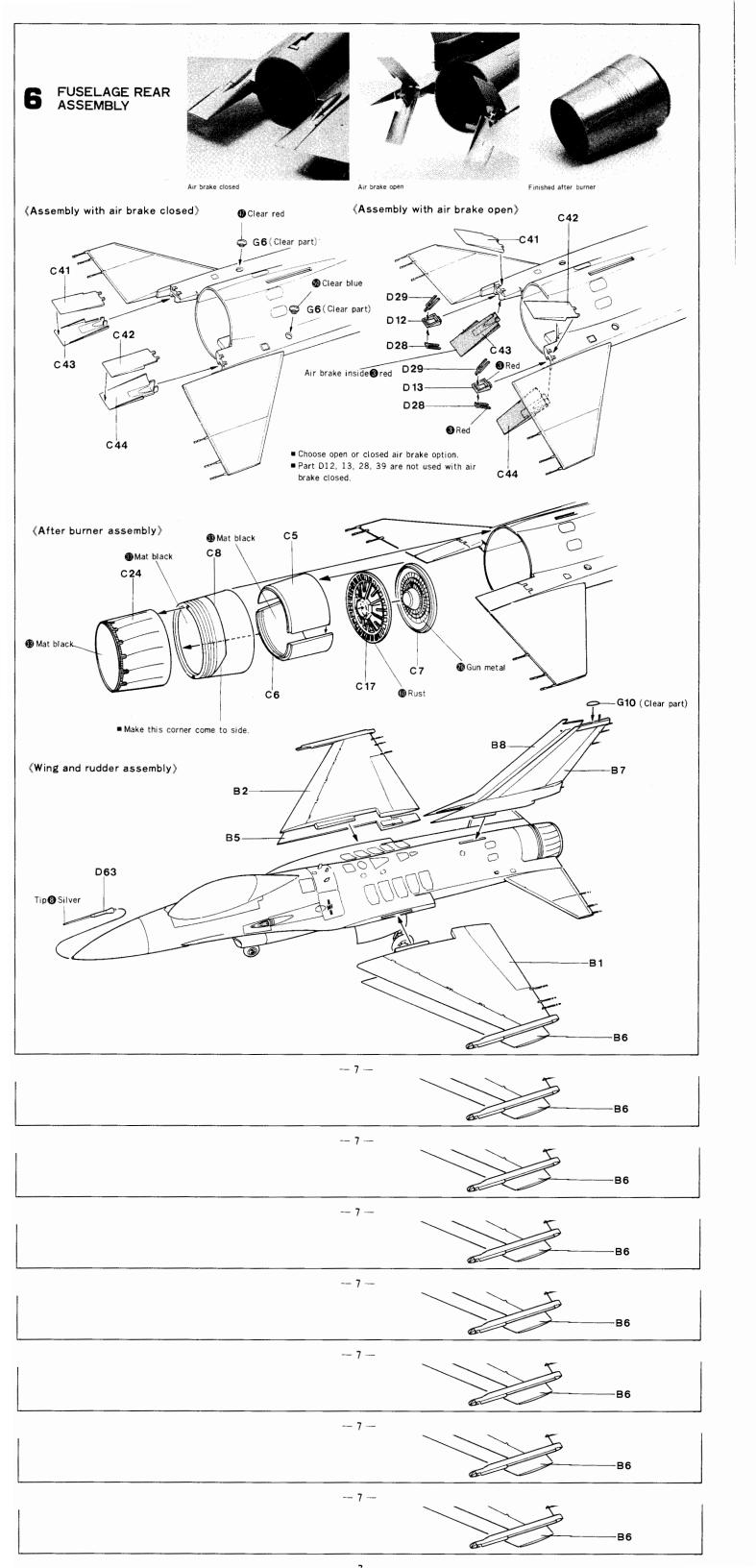


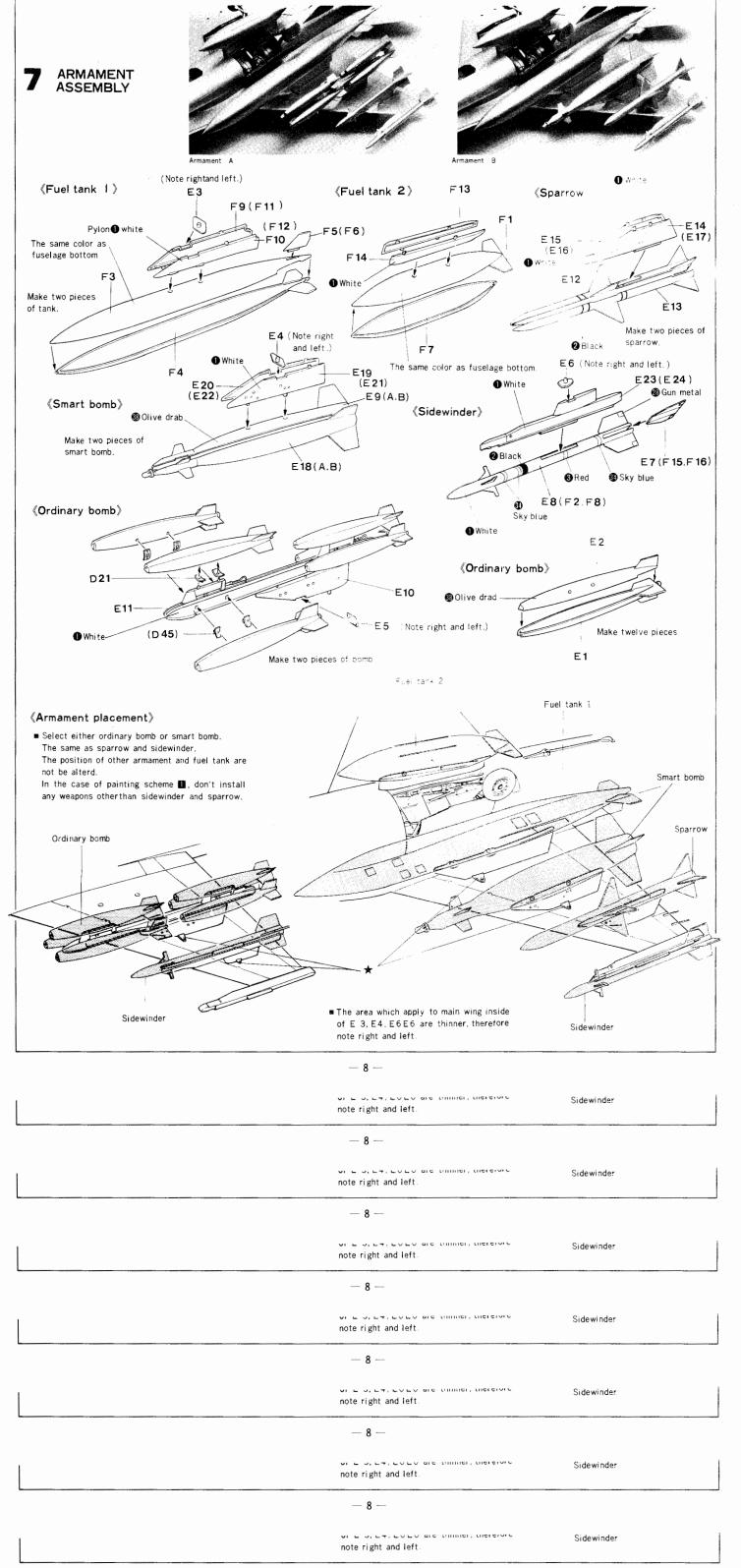


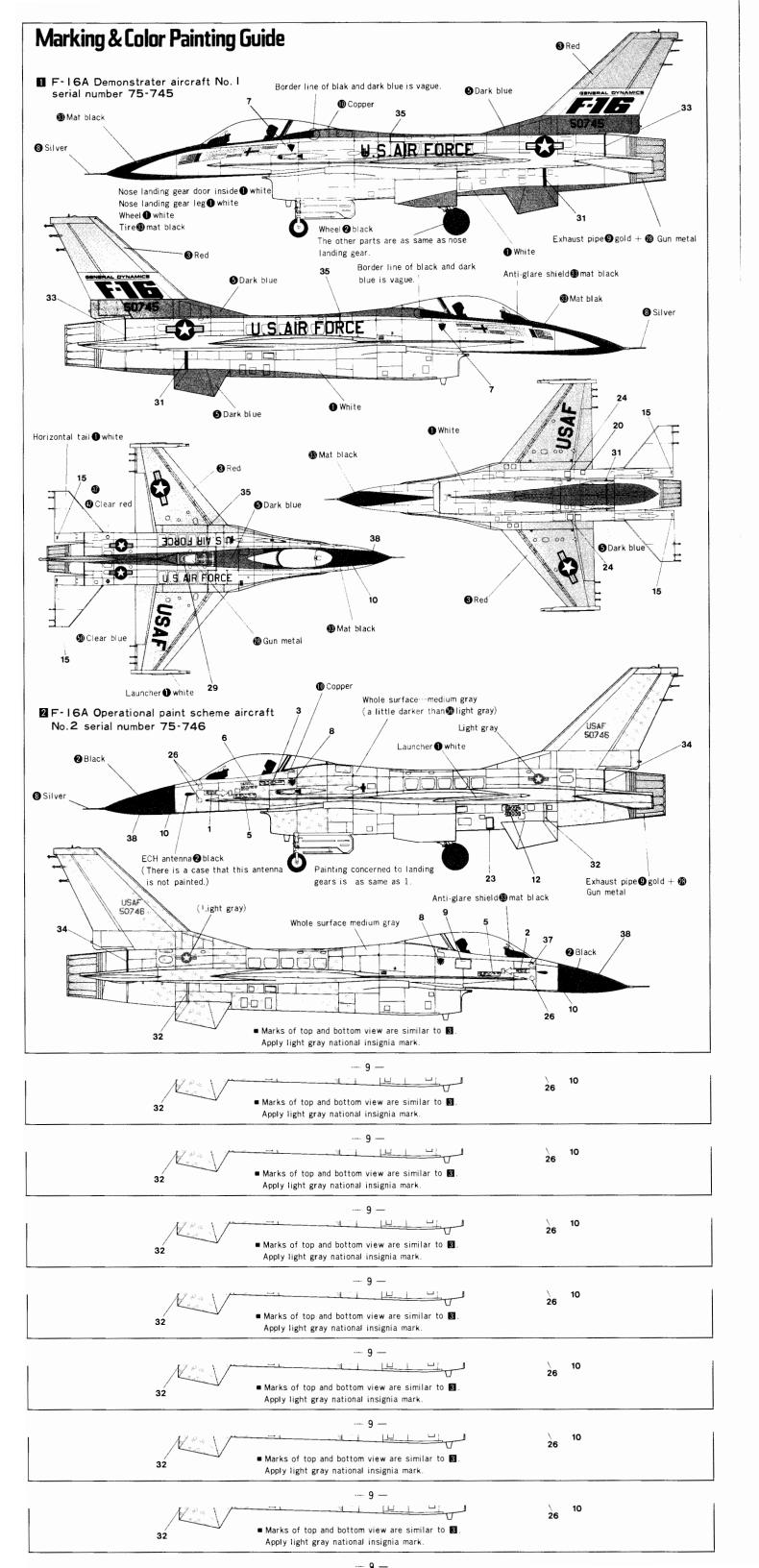


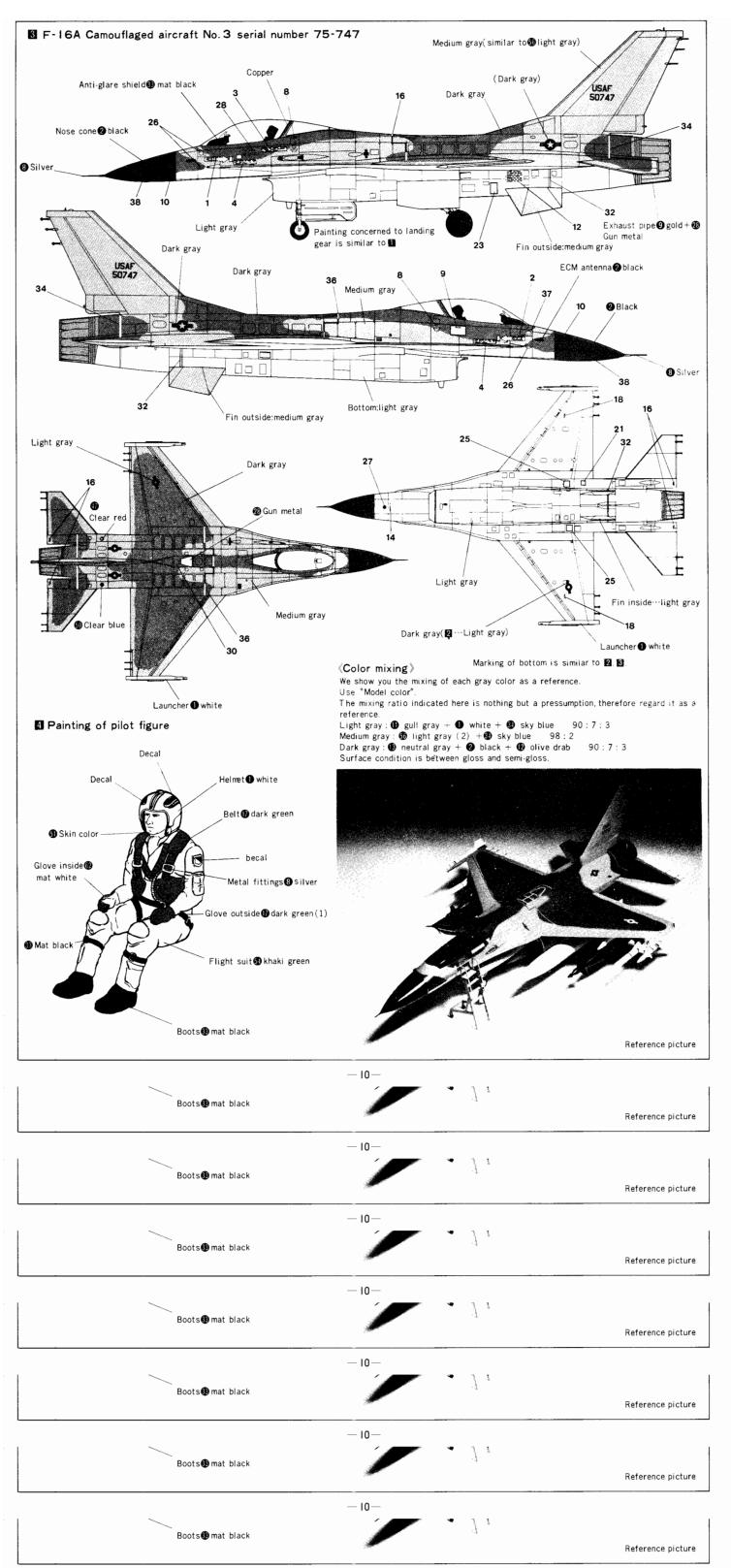


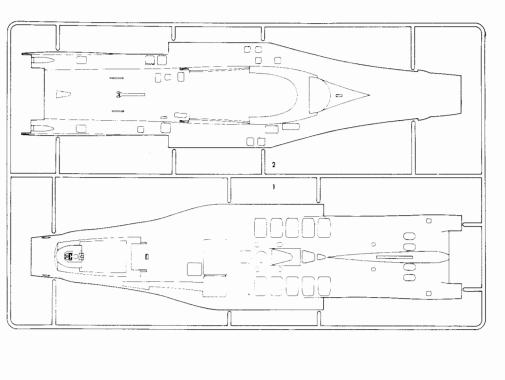


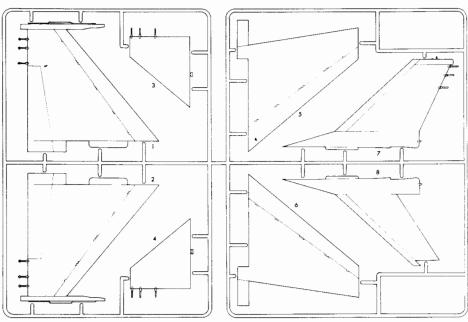


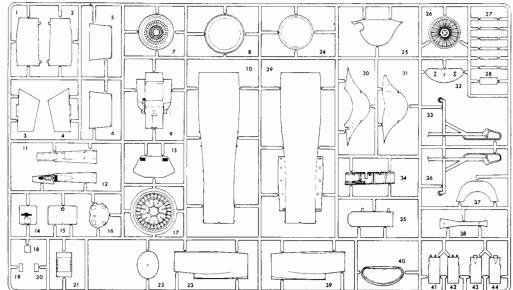












PARTS NO. & NAME

A-parts

-). Fuselage top
- 2 . Fuselage bottom

B-parts

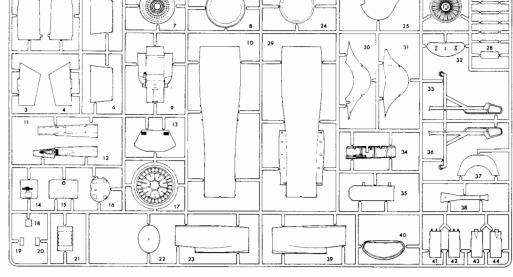
- I. Main wing top(L)
- 2. Main wing top(R) 3. Horizontal tail(R)
- 4 . Horizontal tail(L)
- 5 . Main wing bottom(R)
- 6 . Main wing bottom(L)
- 7. Rudder(L)
- 8 . Rudder(R)

C-parts

- Main gear door(L)
- 2. Main gear door(R)
- 3. Ventral fin(R)
- 4. Ventral fin(L) 5 After burner
- 6 . After burner
- 7. After burner part(B)
- 8. After burner cover
- 9 . Cockpit floor
- Air duct(top)
- II. Vulcan cannon well
- 12. Vulcan cannon access panel 13. Cockpit bulk head
- 14. Vulcan cannon body access panel
- 15. Bullet supplyer access panel
- 16. Radar base
- 17. After burner igniter
- 18. Refueling boom receiver

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- 19. Supplementary air intake
- 20. Supplementary air intake 21. Refueling boom receiver box
- 22. Nose cone
- 23. Air intake section(R)
- 24. Exhaust pipe 25. Bulk head(A)
- 26. Engine turbine fan
- 27. Ladder part(A)
- 28. Ladder part(B)
- 29. Air duct(bottom)
- 30. Bulk head(C)
- 31. Bulk head(B)
- 32. Air intake cover 33. Ladder frame(L)
- 34. Main gear well frame
- 35. Nose gear door
- 36. Ladder frame(R)
- 37. Duct strut bulk head
- Landing gear well cover
 Air intake rear(L)
- 40. Air intake
- 41. Air brake top(L)
- 42. Air brake top(R)
- 43. Air brake bottom(L) 44. Air brake bottom(R)





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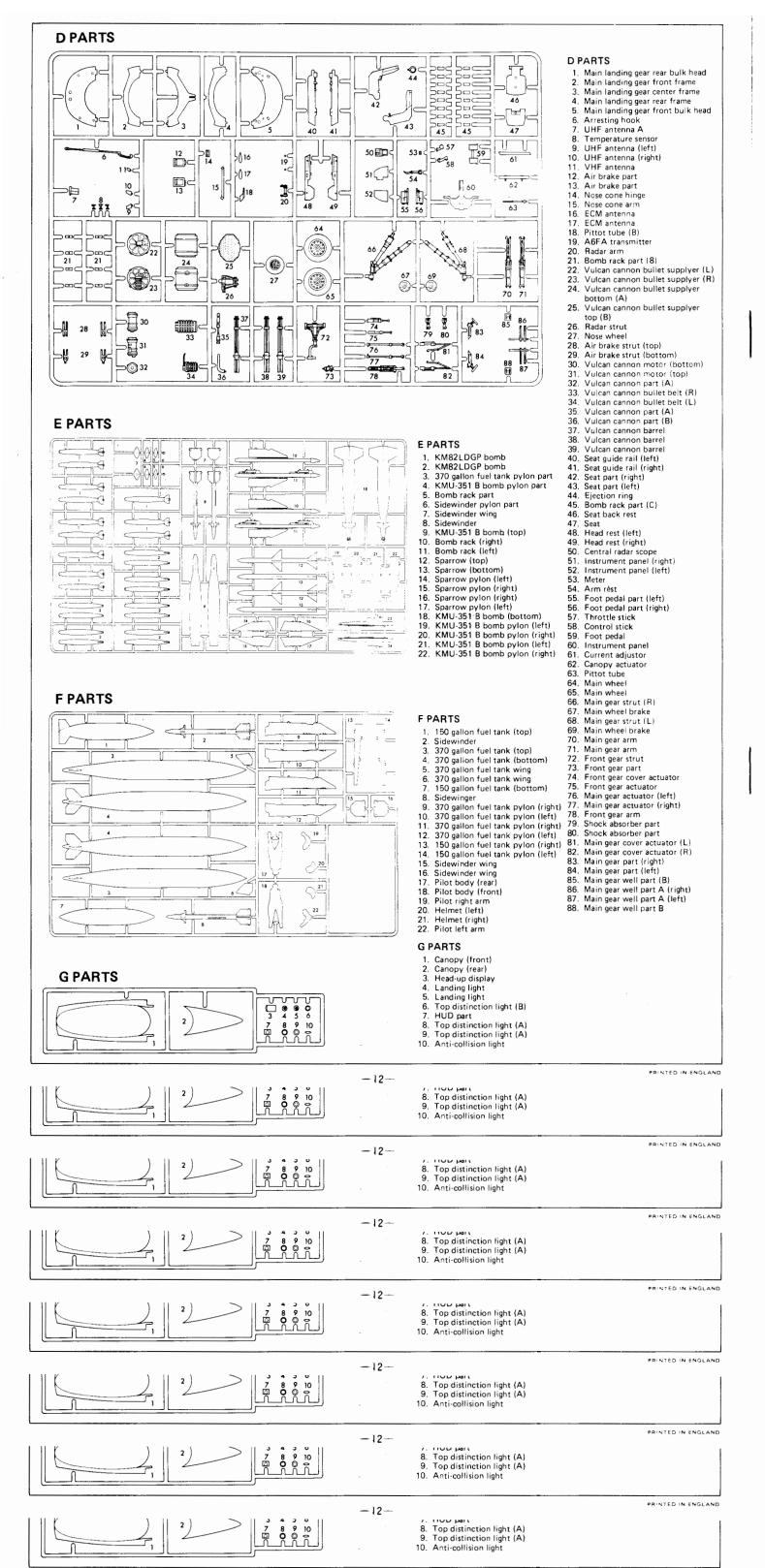












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